

Amendments to the Claims:

This listing of claims will replace all prior versions, and listings, of claims in the application:

Listing of Claims:

1-14. (cancelled)

15. (currently amended) A gas turbine engine fuel injection and combustor system, comprising:

an outer casing extending from an upstream end to a downstream end, an internal space of the downstream end defining a combustion chamber;

5 an annular dome connected to an internal, upstream end of the outer casing;

a radial flow air swirler mounted to a radial flow air swirler housing
~~an internal surface of the annular dome and to an external surface of a fuel~~
~~injector body~~, said swirler providing swirled air to the combustion chamber, said
10 radial flow air swirler housing connected to said annular dome; and

a fuel injector body mounted to an internal surface of the radial flow air swirler housing, said fuel injector body comprising a main circuit fuel nozzle disposed about an outer periphery of said fuel injector body and concentrically disposed about a plurality of axially oriented air swirlers, said air
15 swirlers located within said fuel injector body and concentrically disposed about a pilot circuit fuel nozzle and between said main circuit fuel nozzle and said pilot circuit fuel nozzle, said main circuit fuel nozzle in fluid communication with a main circuit burner fuel through a passageway formed within said fuel injector body, said axially oriented air swirlers formed parallel to a combustion centerline
20 formed within said fuel injector body, said pilot circuit fuel nozzle in fluid communication with a pilot circuit burner fuel through said combustion centerline

~~a plurality of axially oriented air swirlers, an axially located pilot circuit fuel nozzle, and a plurality of radially oriented main circuit fuel nozzles; and~~

25 wherein the annular dome includes a plurality of circumferentially disposed dome cooling nozzles; ~~and~~

~~wherein said fuel injector body incorporates both the pilot circuit and the main circuit fuel nozzles.~~

16. (cancelled)

17. (previously presented) The gas turbine engine fuel injection and combustor system of claim 15, wherein said fuel injector body for the pilot and main circuit fuel nozzles is cylindrical.

18. (cancelled)

19. (previously presented) The gas turbine engine fuel injection and combustor system of claim 15, wherein said pilot circuit fuel nozzle is encircled by the plurality of axially oriented air swirlers.

20. (currently amended) The gas turbine engine fuel injection and combustor system of claim 15, wherein said main circuit fuel nozzles each discharge fuel at ~~an~~ an angle with respect to a combustor assembly centerline axis ~~with respect to a radially extending axial axis of each nozzle~~ into a radial swirler passage.

21. (cancelled)

22. (previously presented) The gas turbine engine fuel injection and combustor system of claim 15, further comprising a can type combustor.

23. (previously presented) The gas turbine engine fuel injection and combustor system of claim 15, further comprising an annular type combustor.

24. (previously presented) The gas turbine engine fuel injection and combustor system of claim 15, wherein the system is capable of utilizing aviation fuel.

25. (currently amended) A gas turbine engine fuel injection and combustor system, comprising:

an outer casing extending from an upstream end to a downstream end, an internal space of the downstream end defining a combustion chamber;

5 an annular dome connected to an internal, upstream end of the outer casing;

a radial flow air swirler mounted to a radial flow air swirler housing ~~an internal surface of the annular dome and to an external surface of a fuel injector body~~, said swirler providing swirled air to the combustion chamber, said radial flow air swirler housing connected to said annular dome; and

10 a fuel injector body mounted to an internal surface of the radial flow air swirler housing, said fuel injector body comprising a main circuit fuel nozzle disposed about an outer periphery of said fuel injector body and concentrically disposed about a plurality of axially oriented air swirlers, said air swirlers located within said fuel injector body and concentrically disposed about a pilot circuit fuel nozzle and between said main circuit fuel nozzle and said pilot circuit fuel nozzle, said main circuit fuel nozzle in fluid communication with a main circuit burner fuel through a passageway formed within said fuel injector body, said axially oriented air swirlers formed parallel to a combustion centerline formed within said fuel injector body, and said pilot circuit fuel nozzle in fluid communication with a pilot circuit burner fuel through said combustion centerline a plurality of axially oriented air swirlers, an axially located pilot circuit fuel nozzle, and a plurality of radially oriented main circuit fuel nozzles;

25 ~~wherein said fuel injector body incorporates both the pilot circuit
and the main circuit fuel nozzles.~~

26. (cancelled)

27. (previously presented) The gas turbine engine fuel injection and combustor system of claim 25, wherein said fuel injector body for the pilot and main circuit fuel nozzles is cylindrical.

28. (cancelled)

29. (previously presented) The gas turbine engine fuel injection and combustor system of claim 25, wherein said pilot circuit fuel nozzle is encircled by the plurality of axially oriented air swirlers.

30. (currently amended) The gas turbine engine fuel injection and combustor system of claim 25, wherein said main circuit fuel nozzles each discharge fuel at an an angle with respect to a combustor assembly centerline axis ~~with respect to a radially extending axial axis of each nozzle~~ into a radial swirler passage.

31. (previously presented) The gas turbine engine fuel injection and combustor system of claim 25, further comprising a can type combustor.

32. (previously presented) The gas turbine engine fuel injection and combustor system of claim 25, further comprising an annular type combustor.

33. (previously presented) The gas turbine engine fuel injection and combustor system of claim 25, wherein the system is capable of utilizing aviation fuel.

34-42. (cancelled)